Milestone Review Flysheet

Institution

University of South Florida

Vehicle Properties		
Total Length (in)	138	
Diameter (in)	4	
Gross Lift Off Weigh (Ib)	23.2	
Airframe Material	G12 Fiberglass	
Fin Material	G10 Fiberglass	
Drag	62 lbf	

Stability Analysis		
Center of Pressure (in from nose)	114 in	
Center of Gravity (in from nose)	92.512 in	
Static Stability Margin	5.38	
Static Stability Margin (off launch rail)	5.38	
Thrust-to-Weight Ratio	8.93	
Rail Size and Length (in)	121	
Rail Exit Velocity	82.3 ft/s	

Recovery System Properties				
Dogue Parachute				
Manufactu	irer/Model	SkyAngle		
Si	ze		6.3 sq ft	
Altitude at Deployment (ft)		Apogee		
Velocity at Deployment (ft/s)		()	
Terminal Velocity (ft/s)		63.04		
Recovery Harness Material		Tubular Nylon		
Harness Size/Thickness (in)		1		
Recovery Harness Length (ft)		34.5		
Connectic Harness/Airframe Interfaces Nosecone/P Altimeter Bay		Connection Nosecone/Pa Altimeter Bay	n between eye b Iyload Bay Bulkh fastened to the	oolts on the lead and Fore Fore airframe.
Kinetic Enerfy	Section 1	Section 2	Section 3	Section 4
of Each Section (Ft-lbs)	190.8	219.745	448.067	

Recovery Electonics			
Altimeter(s)/Timer(s) (Make/Model)	RRC3/Missile Works		
Redundancy Plan	2 Altimeters wired to redundant seperation charges		
Pad Stay Time (Launch Configuration)	3 Hours		

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Flight Readiness Report

Motor Properties		
Motor Manufacturer	Cesaroni	
Motor Designation	CS L910s	
Max/Average Thrust (lb)	244.165/203.924	
Total Impulse (lbf-s)	1919.21	
Mass Before/After Burn	23.2/17.1 lb	
Liftoff Thrust (lb)	235	

Ascent Analysis		
Maximum Veloxity (ft/s)	746	
Maximum Mach Number	0.67	
Maximum Acceleration (ft/s^2)	298	
Target Apogee (From Simulations)	5280	
Stable Velocity (ft/s)	85	
Distance to Stable Velocity (ft)	11	

Recovery System Properties					
Main Parachute					
Manufactu	irer/Model		SkyAngle		
Siz	ze		56 sq. ft		
Altitu	Altitude at Deployment (ft)		500		
Velocit	ty at Deploymer	ıt (ft/s)	63	63.04	
Terminal Velocity (ft/s)		15.93			
Recovery Harness Material		Tubular Nylon			
Harness Size/Thickness (in)			1		
Recovery Harness Length (ft)		34.5			
Connecti Harness/Airframe Interfaces altimeter Ba rir		Connection altimeter Bay ring	on between eye bolts on Aft and eye bolts on top centering g of the motor mount		
Kinetic Enerfy	Section 1	Section 2	Section 3	Section 4	
of Each Section (Ft-lbs)	12.184	14.032	26.612		

Recovery Electonics		
Rocket Locators (Make/Model)	TeleGPS/Apogee	
Transmitting Frequencies	100kHz Band starting at 434.550 MHz	
Black Powder Mass Drogue Chute (grams)	3.5 g	
Black Powder Mass Main Chute (grams)	4 g	

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	Autonomous Ground Support Equipment (MAV Teams Only)		
	Overview		
Capture Mechanism	The robotic arm will be attached to the base of a rover mechanism. After the payload has been approached a camera on the end of the robotic gripper will determine the payload location and orient itself for capture. Upon capture the rover will return to a predefined base and confirm payload location and orientation at a static camera attached to the base of the rail.		
	Overview		
Container Mechanism	The container mechanism for the AGSE is a linear actuator system, with an attached payload sled that can be closed or opened with radio signalling.		
	Overview		
Launch Rail Mechanism	The launch rail will be lifted by a linear actuator system. The extended rod will revent slippage and will be place to geometrically allow 5 degrees off vertical at full extension.		
	Overview		
lgniter Installation Mechanism	The igniter will be installed in a linear actuator attached to the baseplate of the AGSE rail. The igniter will be kept straight by a guide hole in the blast plate and directed upwards by the linear actuator after the launch rail is in position.		

	Payload			
	Overview			
Payload 1	The AGSE payload will be a sealed PVC pipe 3" in length and 3/4" in diameter filled with sand in order to weigh 4 oz.			
	Overview			
Payload 2				

	Test Plans, Status, and Results		
Ejection Charge Tests	The ejection charge tests precede each launch with programming of altimeter and test fires.		
Sub-scale Test Flights	The subscale launch was a success with an achieved altitude of 4092 feet.		
Full-scale Test Flights			

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Additional Comm	nents	